

## Article

# Dynamic Analysis of a Fiber-Reinforced Composite Beam under a Moving Load by the Ritz Method

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**Abstract:** This paper presents the dynamic responses of a fiber-reinforced composite beam under a moving load. The Timoshenko beam theory was employed to analyze the kinematics of the composite beam. The constitutive equations for motion were obtained by utilizing the Lagrange procedure. The Ritz method with polynomial functions was employed to solve the resulting equations in conjunction with the Newmark average acceleration method (NAAM). The influence of fiber orientation angle, volume fraction, and velocity of the moving load on the dynamic responses of the fiber-reinforced nonhomogeneous beam is presented and discussed.

**Keywords:** moving load problems; fiber-reinforced composite materials; Timoshenko beams; Ritz method

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## 1. Introduction

Conventional engineering materials may be classified as metals, polymers, ceramics, and composites. Metals have high rigidity, ductility, mechanical strength, and thermal stability. They are also exceedingly good conductors of electricity and heat. Polymers have become one of the widely used engineering materials due to their lower density, easy machinability, and corrosion resistance compared to other materials. Ceramics contain strong covalent bonds and have almost zero thermal and electrical conductivities and extremely good thermal stability and hardness.

A composite material is any solid consisting of at least two components contained in separate phases. Wood is a natural composition of cellulose fibers in a lignin matrix. Another well-known example of man-made composite material is reinforced concrete. Steel and concrete maintain their individual characteristics in the resulting composite structure. However, because they work in harmony, the steel carries tension loads, and the concrete withstands compression loads. The main advantages of composite materials include excellent strength-to-weight and stiffness-to-weight ratios. Additionally, composite materials may have excellent resistance to corrosion.

Fiber-reinforced composite (FRC) structures are used in various engineering applications, for example, airplanes, machine, marine, and civil engineering projects, because of their higher strength-to-weight ratios, greater lightweight, and better ductile properties [1–9]. The main practical applications of FRC structures comprise molded car panels, helicopter and wind turbine blades, tennis rackets, ski-poles, and prosthesis. More details can be found in [10,11]. Issues associated with a moving load are very important in FRC structures used in bridges, roads, railways structures. With dynamically moving loads,

the mechanical behavior of structures varies significantly [12–18]. Therefore, understanding the dynamical behavior of FRC structures is crucial for their design. Some works on the mechanical analysis of FRC beams are reported below.

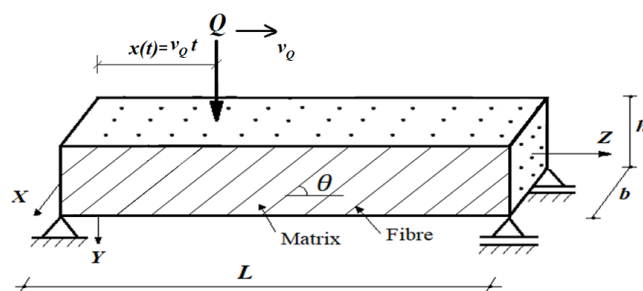
A dynamic analysis of beams reinforced with fibers was performed by Teoh and Huang [19]. Bending and stability responses of thin-walled beams reinforced with cast polyamide, carbon, and glass fibers were perused by Eksi et al. [20]. The vibrational response of a laminated beam with a SiC particle-reinforced Al6082 matrix in a magnetorheological fluid core was examined [21]. Large-amplitude vibration analyses of laminated imperfect FRC beams were performed by Jin et al. [22]. Post-buckling and large deflection analyses of an FRC beam were made with the finite element method by Akbas [23–25]. In addition, the mechanical responses of composite beams reinforced with carbon fiber have attracted the attention of researchers [26–29].

On the other hand, carbon nanotubes (CNTs) have also been used as a reinforcement material. Due to the extraordinary properties of CNTs, many studies have been performed on their mechanical characteristics [30–33]. The bending and buckling responses of a nanocomposite beam reinforced with SWCNT were comprehensively examined [34]. Ke et al. [35] performed a large-amplitude vibration analysis of nonhomogeneous CNT-reinforced beams on the basis of the Timoshenko beam theory. Yas and Samadi [36] perused the dynamic and stability responses of CNT-reinforced composite beams surrounded by an elastic medium by GDQM. More details about nanocomposite beams reinforced with CNTs can be found in other related works [37–49]. Moreover, the vibration and buckling behaviors of nanocomposite beams reinforced with graphene nanoplatelets were studied [50–53].

As seen above, there are a number of studies on the mechanical analysis of reinforced composite beams in the scientific literature. However, works about the dynamic analysis of FRC beams under a moving load are very limited. To the best of the authors' knowledge, the dynamical response of fiber-reinforced composite beams subjected to a moving load using the Newmark average acceleration and Ritz methods was examined for the first time in a simply supported boundary condition. In the present study, dynamically moving issues related to an FRC beam are investigated using the Timoshenko beam theory and Ritz method. The governing equations for a reinforced composite beam in dynamic analysis were obtained by using the Lagrange procedure. NAAM was utilized to solve the forced vibration problem in time. The influence of fiber orientation angles, volume fraction, and velocity of the moving load on the dynamical behavior of the FRC beam is presented.

## 2. Materials and Methods

In Figure 1, a simply supported FRC beam with length  $L$ , height  $h$ , and width  $b$  under a moving load is shown.  $Q$  and  $v_Q$  represent the magnitude and the constant velocity of the load, respectively.



**Figure 1.** A simply supported FRC beam under a moving load.

The axial ( $\epsilon_z$ ) and shear ( $\gamma_{zy}$ ) strains can be written based on the first-order shear deformation beam theory as

$$\varepsilon_z = \frac{\partial u_0}{\partial z} - Y \frac{\partial \varnothing}{\partial z} \quad (1a)$$

$$\gamma_{zy} = \frac{\partial u_0}{\partial y} + \frac{\partial v_0}{\partial z} \quad (1b)$$

where  $u_0$ ,  $v_0$ , and  $\varnothing$  denote axial and vertical displacements and rotation, respectively. The constitutive relation is presented as follows

$$\begin{Bmatrix} \sigma_z \\ \sigma_{zy} \end{Bmatrix} = \begin{bmatrix} \bar{R}_{11} & \bar{R}_{16} \\ \bar{R}_{16} & \bar{R}_{66} \end{bmatrix} \begin{Bmatrix} \varepsilon_z \\ \gamma_{zy} \end{Bmatrix} \quad (2)$$

where  $\bar{R}_{ij}$  represents the transformed components of the reduced constitutive tensor. They can be expressed for orthotropic materials as

$$\bar{R}_{11} = Q_{11}c^4 + 2(Q_{12} + 2Q_{66})c^2s^2 + Q_{22}s^4 \quad (3a)$$

$$\bar{R}_{12} = (Q_{11} + Q_{22} - 4Q_{66})s^2c^2 + Q_{12}(c^4 + s^4) \quad (3b)$$

$$\bar{R}_{16} = (Q_{11} - Q_{12} - 2Q_{66})sc^3 + (Q_{12} - Q_{22} + 2Q_{66})s^3c \quad (3c)$$

$$\bar{R}_{22} = Q_{11}s^4 + 2(Q_{12} + 2Q_{66})s^2c^2 + Q_{22}c^4 \quad (3d)$$

$$\bar{R}_{26} = (Q_{11} - Q_{12} - 2Q_{66})s^3c + (Q_{12} - Q_{22} + 2Q_{66})sc^3 \quad (3e)$$

$$\bar{R}_{66} = (Q_{11} + Q_{22} - 2Q_{12} - 2Q_{66})s^2c^2 + Q_{66}(s^4 + c^4) \quad (3f)$$

where  $c = \cos \theta$ , and  $s = \sin \theta$ .  $\theta$  specifies the fiber orientation angle, and  $Q_{ij}$  can be defined as

$$Q_{11} = \frac{E_1}{1 - \nu_{12}\nu_{21}}, Q_{22} = \frac{E_2}{1 - \nu_{12}\nu_{21}} \quad (4a)$$

$$Q_{12} = Q_{21} = \frac{\nu_{12}E_2}{1 - \nu_{12}\nu_{21}} = \frac{\nu_{21}E_1}{1 - \nu_{12}\nu_{21}} \quad (4b)$$

$$Q_{66} = G_{12} \quad (4c)$$

where  $E_1$  and  $E_2$  are, respectively, the elastic modulus in the Z and Y directions,  $\nu_{12}$  and  $\nu_{21}$  are Poisson's ratios, and  $G_{12}$  is the shear modulus in the ZY plane. The mechanical properties of composite materials can be calculated according to the relations in Eqs. (5a)-(5e) [54]

$$E_1 = E_f V_f + E_m (1 - V_f) \quad (5a)$$

$$E_2 = E_m \left[ \frac{E_f + E_m + (E_f - E_m)V_f}{E_f + E_m - (E_f - E_m)V_f} \right] \quad (5b)$$

$$\nu_{12} = \nu_f V_f + \nu_m (1 - V_f) \quad (5c)$$

$$G_{12} = G_m \left[ \frac{G_f + G_m + (G_f - G_m)V_f}{G_f + G_m - (G_f - G_m)V_f} \right] \quad (5d)$$

$$\rho = \rho_f V_f + \rho_m (1 - V_f) \quad (5e)$$

where  $f$  and  $m$  indicate the fiber and matrix, respectively, and  $V_f$  is the volume fraction of the fiber. Also,  $E$ ,  $G$ ,  $\nu$ , and  $\rho$  denote the material's properties, i.e., Young's modulus, shear modulus, Poisson's ratio, and mass density, respectively.

The strain energy ( $U_s$ ), kinetic energy ( $T$ ), and potential energy of the external loads ( $U_e$ ) can be expressed as

$$U_s = \frac{1}{2} \int_0^L \left[ A_0 \left( \frac{\partial u_0}{\partial z} \right)^2 + A_2 \left( \frac{\partial \phi}{\partial z} \right)^2 \right] dz + \frac{1}{2} \int_0^L K_s B_0 \left[ \left( \frac{\partial v_0}{\partial z} \right)^2 - 2 \frac{\partial v_0}{\partial z} \phi + \phi^2 \right] dz \quad (6a)$$

$$T = \frac{1}{2} \int_0^L \left( I_0 \left( \frac{\partial u_0}{\partial t} \right)^2 + I_2 \left( \frac{\partial \phi}{\partial t} \right)^2 + I_0 \left( \frac{\partial v_0}{\partial t} \right)^2 \right) dz \quad (6b)$$

where  $K_s$  is a shear correction factor, chosen as 5/6 in the present study and

$$A_0 = \bar{R}_{11} b h, A_2 = \bar{R}_{11} b h^3 / 12, B_0 = \bar{R}_{66} b h, I_0 = \rho b h, I_2 = \rho b h^3 / 12 \quad (7)$$

The Lagrangian function can be written as

$$I = T - (U_i + U_e) \quad (8)$$

An approximate solution can be achieved as a series of  $n$  terms of the following form, according to the Ritz method

$$u_0(z, t) = \sum_{i=1}^{\infty} d_n(t) \varphi_n(z) \quad (9a)$$

$$v_0(z, t) = \sum_{i=1}^{\infty} e_n(t) \chi_n(z) \quad (9b)$$

$$\phi(z, t) = \sum_{i=1}^{\infty} f_n(t) \psi_n(z) \quad (9c)$$

where  $d_n$ ,  $e_n$ , and  $f_n$  are the unknown time-dependent generalized coordinates, and  $\varphi_n(z)$ ,  $\chi_n(z)$ , and  $\psi_n(z)$  are the admissible functions dependent on the boundary conditions.  $\varphi_n(z)$ ,  $\chi_n(z)$ , and  $\psi_n(z)$  can be written, for a simply supported beam, as

$$\varphi_n(z) = z^n \quad (10a)$$

$$\chi_n(z) = (L - z)z^n \quad (10b)$$

$$\psi_n(z) = z^{(n-1)} \quad (10c)$$

where  $n$  states the order of polynomials in the approximate functions.

Substituting Equations (9a)–(9c) into Equations (6a) and (6b) and then implementing the Lagrange's equation yields Equation (11)

$$\frac{\partial I}{\partial r_n} - \frac{\partial}{\partial t} \frac{\partial I}{\partial \dot{r}_n} = 0 \quad (11)$$

where  $r_n$  represents  $d_n$ ,  $e_n$ , and  $f_n$ . The equation of motion is achieved as

$$[K]\{\ddot{r}(t)\} + [M]\{\ddot{r}(t)\} = \{F(t)\} \quad (12)$$

The components of these matrices and vector are presented below

$$[K] = \begin{bmatrix} K_{11} & K_{12} & K_{13} \\ K_{21} & K_{22} & K_{23} \\ K_{31} & K_{32} & K_{33} \end{bmatrix} \quad (13)$$

where

$$K_{11} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L A_0 \frac{\partial \varphi_i}{\partial z} \frac{\partial \varphi_j}{\partial z} dz, K_{12} = 0, K_{13} = 0 \quad (14a, b, c)$$

$$K_{21} = 0, K_{22} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L K_s B_0 \frac{\partial \chi_i}{\partial z} \frac{\partial \chi_j}{\partial z} dz, K_{23} = - \sum_{i=1}^n \sum_{j=1}^n \int_0^L K_s B_0 \frac{\partial \chi_i}{\partial z} \psi_j dz \quad (14d, e, f)$$

$$K_{31} = 0, K_{32} = - \sum_{i=1}^n \sum_{j=1}^n \int_0^L K_s B_0 \psi_i \frac{\partial \chi_j}{\partial z} dz \quad (14g, h)$$

$$K_{33} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L A_2 \frac{\partial \psi_i}{\partial z} \frac{\partial \psi_j}{\partial z} dz + \sum_{i=1}^n \sum_{j=1}^n \int_0^L K_s B_0 \psi_i \psi_j dz \quad (14i)$$

$$[\mathcal{M}] = \begin{bmatrix} \mathcal{M}_{11} & \mathcal{M}_{12} & \mathcal{M}_{13} \\ \mathcal{M}_{21} & \mathcal{M}_{22} & \mathcal{M}_{23} \\ \mathcal{M}_{31} & \mathcal{M}_{32} & \mathcal{M}_{33} \end{bmatrix} \quad (15)$$

in which

$$\mathcal{M}_{11} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L I_0 \varphi_i \varphi_j dz, \quad \mathcal{M}_{12} = 0, \quad \mathcal{M}_{13} = 0 \quad (16a, b, c)$$

$$\mathcal{M}_{21} = 0, \quad \mathcal{M}_{22} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L I_0 \chi_i \chi_j dz, \quad \mathcal{M}_{23} = 0 \quad (16d, e, f)$$

$$\mathcal{M}_{31} = 0, \quad \mathcal{M}_{32} = 0 \quad (16g, h)$$

$$\mathcal{M}_{33} = \sum_{i=1}^n \sum_{j=1}^n \int_0^L I_2 \psi_i \psi_j dz \quad (16i)$$

$$\{F(t)\} = Q \chi_j(v_Q t) \quad 0 \leq t \leq L/v_Q \quad (17)$$

Equation (12) can be solved within the time domain by NAAM. The procedure of NAAM is presented as follows

$$[\bar{K}(t)]\{d_n\}_{r+1} = \{\bar{F}(t)\} \quad (18)$$

where

$$[\bar{K}(t)] = [K] + a_0[\mathcal{M}] \quad (19)$$

$$\{\bar{F}(t)\} = \{\bar{F}(t)\}_{r+1} + [\mathcal{M}] \left( a_0\{d_n\}_r + a_1\{\dot{d}_n\}_r + a_2\{\ddot{d}_n\}_r \right) \quad (20)$$

where

$$a_0 = \frac{1}{\beta \Delta t^2}, \quad a_1 = \frac{1}{\beta \Delta t}, \quad a_2 = \frac{1-2\beta}{\beta}, \quad (21)$$

where  $\alpha = 0.5, \beta = 0.25$ . After evaluating  $\{d_n\}_{r+1}$  at time  $t_{r+1} = t_r + \Delta t$ , the acceleration and velocity vectors can be determined by

$$\{\ddot{d}_n\}_{j+1} = a_0(\{d_n\}_{r+1} - \{d_n\}_r) - a_1\{\dot{d}_n\}_r - a_2\{\ddot{d}_n\}_r \quad (22)$$

$$\{\dot{d}_n\}_{r+1} = \{\dot{d}_n\}_r + a_3\{\ddot{d}_n\}_r + a_4\{\ddot{d}_n\}_{r+1} \quad (23)$$

where  $a_3 = (1 - \alpha)\Delta t$ , and  $a_4 = \alpha\Delta t$ .

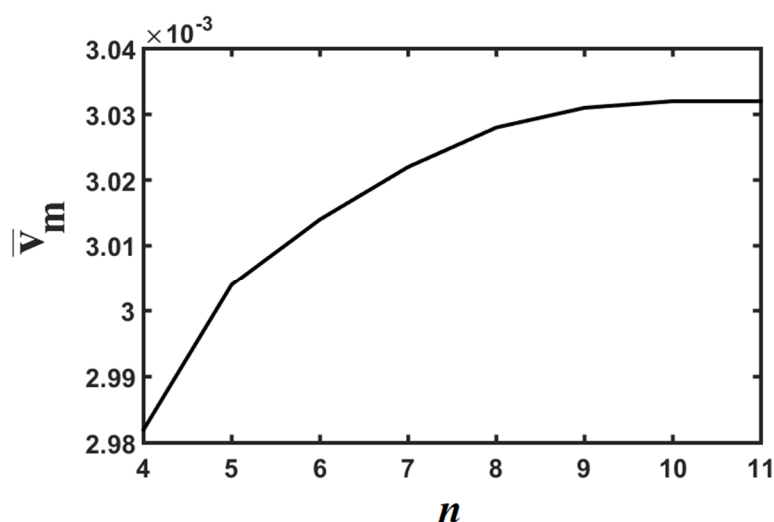
The dimensionless transversal displacement and time are respectively presented as follows

$$\bar{v} = \frac{v}{L}, \quad t^* = \frac{t}{L} v_Q \quad (24)$$

### 3. Numerical Results

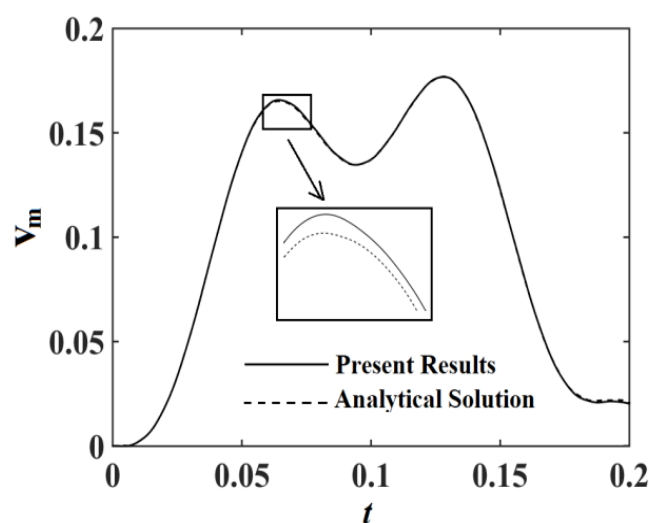
In this section, dynamical displacements of the FRC simply supported beam under a moving load are presented and discussed according to different material and load parameters. In the numerical examples, the materials of the beams were selected as consisting of a graphite fiber-reinforced polyamide composite, and its parameters were as follows [55]:  $E_m = 2.756$  GPa,  $E_f = 275.6$  GPa,  $G_m = 1.036$  GPa,  $G_f = 114.8$  GPa,  $\nu_m = 0.33$ ,  $\nu_f = 0.2$ ,  $\rho_m = 1600$  kg/m<sup>3</sup>,  $\rho_f = 1900$  kg/m<sup>3</sup>. The geometry properties of the beam were selected as  $b = h = 0.1$  m, and  $L = 10h$ . The amplitude of the dynamic load was selected as  $Q_0 = 1$  kN.

Firstly, a convergence study was performed to determine the suitable number of polynomials, as shown in Figure 2. The maximum dimensionless dynamic displacements at midspan of the FRC beam under a moving load were calculated for  $V_f = 0.3$ ,  $\theta = 10^\circ$ ,  $V_Q = 10$  m/s. It Figure 2, it can be observed that the number of polynomials has a crucial effect on the convergence of the dynamic displacements. It is also apparent that an increase in the number of polynomials yields an increase in the maximum dynamic displacements and the divergence between the maximum dynamic displacements diminishes for a higher number of polynomials, i.e., for  $n > 9$ . Consequently, the number of polynomials is chosen as 10 in the numerical results.



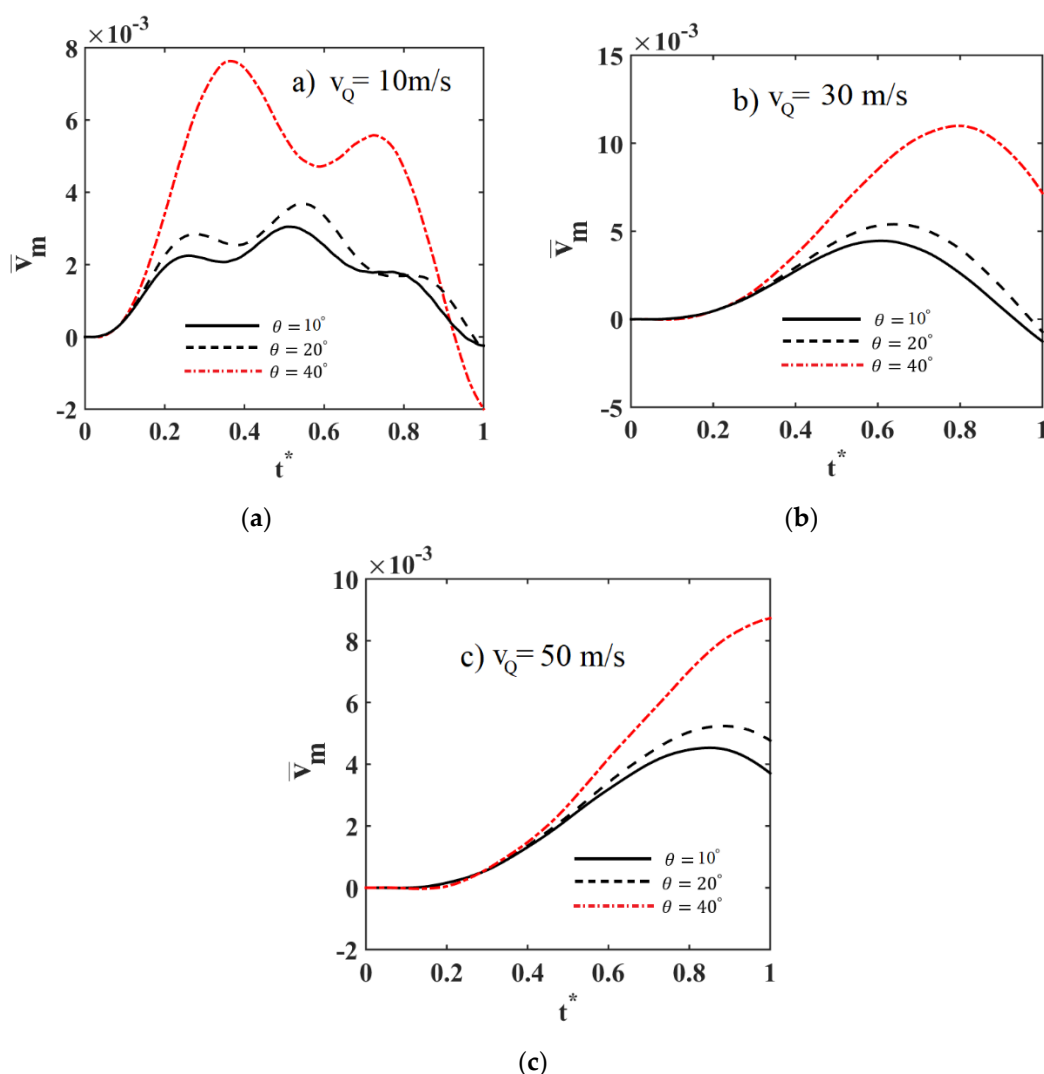
**Figure 2.** Convergence of maximum dimensionless dynamic displacements ( $\bar{v}_m$ ) of the FRC beam under a moving load for  $V_f = 0.3$ ,  $\theta = 10^\circ$ ,  $V_Q = 10$  m/s.

Then, in order to demonstrate the accuracy and validity of the present analysis, the results were compared with those of a previous work available in the literature. For this, the dynamic lateral displacements at midspan of a homogeneous and isotropic beam under a moving load were compared with [56] in Figure 3. For a validation purpose, the material parameters were chosen as  $E = 210$  GPa,  $\rho = 7800$  kg/m<sup>3</sup>,  $\nu = 0.3$ ,  $v_f = 30$  m/s. As shown in this figure, it is clear that our findings are in agreement with other established results.



**Figure 3.** Dynamic lateral displacements at midspan ( $v_m$ ) of a homogeneous and isotropic beam under a moving load.

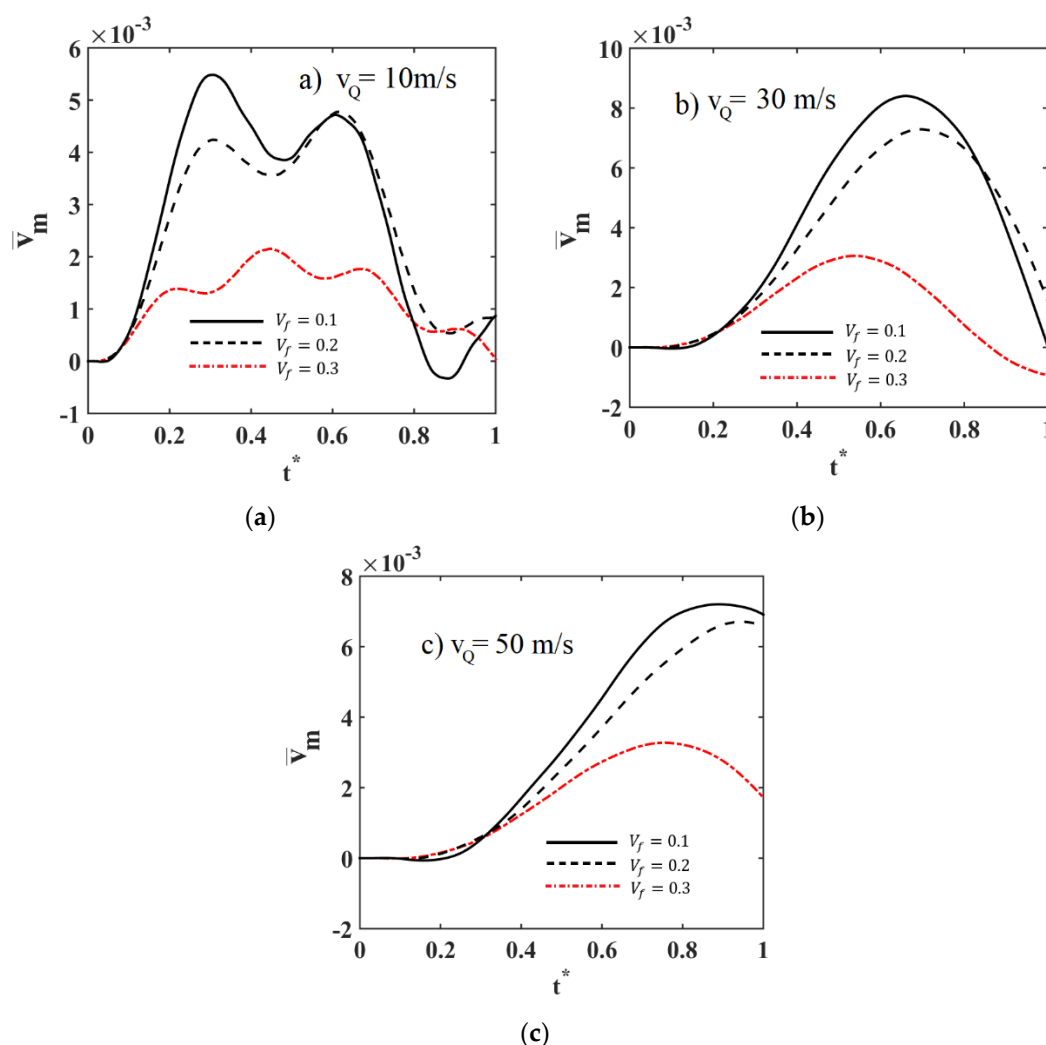
In Figure 4, the effects of the fiber orientation angle ( $\theta$ ) on the dimensionless lateral dynamical displacements at midspan ( $\bar{v}_m$ ) of FRC beams are presented for the different values of  $\theta$  and velocity of loads in the dimensionless time history for  $V_f = 0.3$ . The dimensionless displacements ( $\bar{v}$ ) and time ( $t^*$ ) quantities defined in Equation (24) were used in the time history graphs. As seen in Figure 4, the dynamical lateral dimensionless displacements of the FRC beam increased with an increment in  $\theta$ . The bending rigidity decreased when increasing the fiber orientation angles according to Equation 3, so the displacements increased. In addition, the velocity of the load had an important influence on the effects of the fiber orientation angle. With different values of the velocity of the load, the effects of the fiber orientation angle on the dynamic results changed considerably. For lower values of the velocity of the load, the effect of the fiber orientation angle were larger than for higher load velocities.



**Figure 4.** Time history of dimensionless dynamic displacements at midspan of the FRC beam under a moving load for different values of the fiber orientation angle ( $\theta$ ).

The influences of  $V_f$  and  $V_Q$  on the dimensionless dynamic displacements of a simply supported FRC beam are depicted within time history in Figure 5 for  $\theta = 30^\circ$ . It is seen from Figure 5 that increasing volume fraction of the fiber decreased the dynamical displacements, as expected. It is also evident that the dimensionless dynamic displacements evaluated for  $V_f = 0.1$  and  $V_f = 0.2$  were close to each other, while those for  $V_f = 0.3$  were

very different. In addition, the effects of the volume fraction of fiber on the dynamic responses varied considerably for different values of load's velocity. For different values of  $V_Q$ , the behavior of the fibers in the dynamic response changed. For example, for  $V_Q = 10$  m/s and  $t^* = 1$ , the effects of the volume fraction of fiber were very different from those for other  $V_Q$  and  $t^*$ . This revealed that  $V_Q$  has an important effect on the dynamic responses of FRC beams.



**Figure 5.** Time history of dimensionless dynamic displacements at midspan of the FRC beam under a moving load for different values of the volume fraction of fiber ( $V_f$ ).

#### 4. Conclusions

A dynamic analysis of an FRC simply supported beam under a moving load was carried out based on the Timoshenko beam theory by using the Ritz method. The governing equations in motion were achieved with the aid of the Lagrange procedure. The Newmark average acceleration procedure was employed in the time history. The influence of fiber orientation angle, volume fraction, and load velocity on the dynamic displacements of the FRC beam was investigated in detail. It is concluded from the reported results that the fiber orientation angle and the volume fraction exert an important effect on the dynamic responses of FRC beams under a moving load. With variations in these parameters, the dynamic responses of the FRC beam change significantly. By changing the values of load velocity, the effects of fiber orientation angle and volume fraction on the dynamical responses change considerably.



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